

# CBCS Scheme

USN

1A41SAC006

15AE54

**Fifth Semester B.E. Degree Examination, Dec.2017/Jan.2018**

## Aircraft Structures – I

Time: 3 hrs.

Max. Marks: 80

**Note: Answer any FIVE full questions, choosing one full question from each module.**

### Module-1

- 1
- a. Briefly explain factor of safety in engineering design. (02 Marks)
  - b. A mild steel bracket shown in Fig. Q1(b) is subjected to a pull of 10kN. The bracket has a rectangular cross section whose depth is twice the width. If the allowable stress for material is  $80\text{N/mm}^2$ . Determine the cross section of the bracket. (08 Marks)
  - c. A point in a structural member subjected to plane stress is shown in Fig. Q1(c). Determine the following :
    - i) Normal and tangential stress intensities on a plane inclined at  $40^\circ$
    - ii) Principle stresses and orientations of principal planes
    - iii) Maximum shear stress and direction of plane on which they occur. (06 Marks)

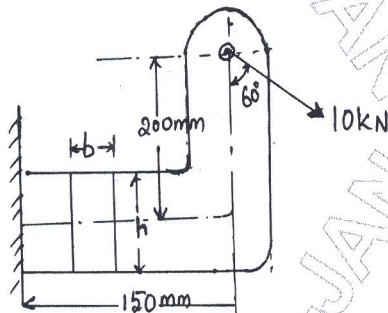


Fig Q1(b)

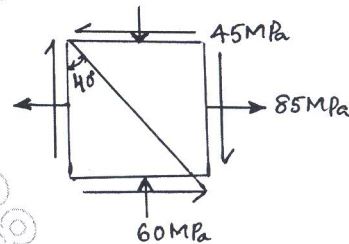


Fig Q1(c)

OR

- 2
- a. A rod of circular section is to sustain a torsional moment of  $300\text{kN-m}$  and bending moment  $200\text{kN-m}$  yield stress for the material is  $353\text{MPa}$  and assuming factor of safety as 3. Determine the diameter of rod as per following theories of failure.
    - i) Maximum shear stress theory
    - ii) Distortion energy theory
    - iii) Maximum strain energy theory (take  $\nu = 0.3$ )
    - iv) Maximum principal stress theory. (08 Marks)
  - b. A bar of Rectangular section is subjected to an axial pull of  $500\text{kN}$ . Calculate its thickness if the allowable stress in the bar is  $200\text{MPa}$  [Fig Q2(b)].

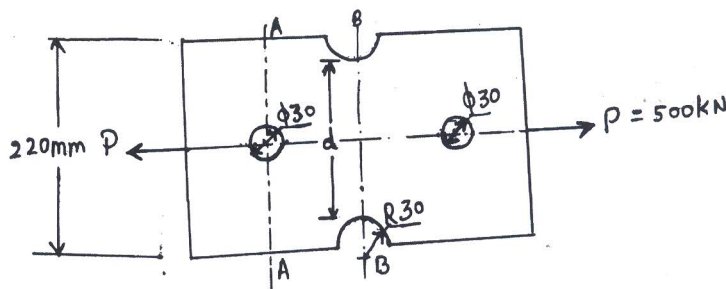


Fig Q2(b)

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(08 Marks)

Important Note : 1. On completing your answers, compulsorily draw diagonal cross lines on the remaining blank pages.  
2. Any revealing of identification, appeal to evaluator and /or equations written eg, 42+8 = 50, will be treated as malpractice.

**Module-2**

- 3 a. A unknown weight falls through 20mm on to a collar rigidly attached to the lower end of a vertical bar 2m long and  $500\text{mm}^2$  section. If the maximum instantaneous extension is 2mm, what is the corresponding stress and value of the unknown weight? Take  $E = 200\text{GPa}$ . (04 Marks)
- b. Explain stress life(S-N) curve for Ferrous material. (06 Marks)
- c. Determine the maximum load for the simply supported beam, cyclically loaded as shown in Fig. 3(c). The ultimate strength is  $700\text{MPa}$ , the yield point in tension is  $520\text{MPa}$  and the endurance limit in reversed bending is  $320\text{MPa}$ . Use a factor of safety of 1.25. The load, size and surface correction factor are 1, 0.75 and 0.9 respectively.

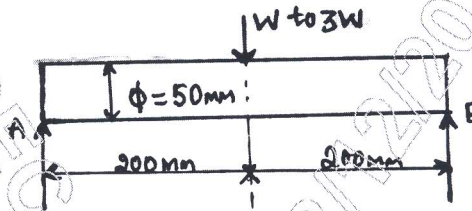


Fig Q3(c)

(06 Marks)

**OR**

- 4 a. Formulate Miner's Rule for cumulative fatigue damage. (04 Marks)
- b. A steel cantilever member as shown in Fig Q4(b), is subjected to a transverse load at its end that varies from  $45\text{N}$  top to  $135\text{N}$  down. An axial load varies from  $110\text{N}$  compression to  $450\text{N}$  tension. Determine the required diameter at the change of section of infinite life using a factor of safety of 2. The strength properties of material are  $\sigma_u = 550\text{MPa}$ ,  $\sigma_y = 470\text{MPa}$  endurance limit from reversed bending test  $\sigma_e = 275\text{MPa}$ . Notch sensitivity index  $q = 1$ .

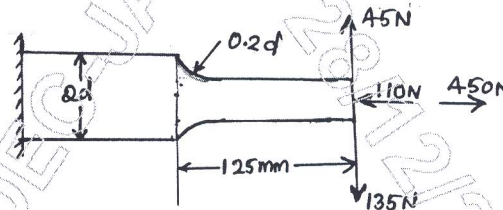


Fig. Q4(b)

(12 Marks)

**Module-3**

- 5 a. With a neat sketch, explain velocity diagram. (08 Marks)
- b. An aircraft having a total weight of  $45\text{kN}$  lands on the deck of a aircraft carrier and is brought to rest by means of a cable engaged by an arrestor hook as shown in Fig. Q5(b). If the deceleration induced by the cable is  $3g$ . Determine the tension  $T$  in the cable, the load on the undercarriage sturt. And shear and axial loads in the fuselage at the section A-A. The weight of the aircraft aft of A-A is  $4.5\text{ kN}$ . Calculate also the length of deck covered by the aircraft before it is brought to rest if the touchdown speed  $25\text{m/s}$ . (08 Marks)

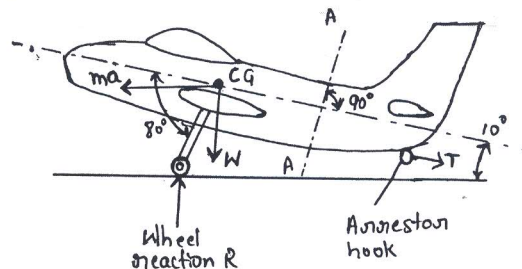


Fig Q5(b)

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OR

- 6 a. Briefly, explain Griffith's theory and derive an expression for stress required for creation of new crack surface. (08 Marks)
- b. List the desirable properties of materials used in aircraft structures. (04 Marks)
- c. Write a short note on Titanium Alloys. (04 Marks)

Module-4

- 7 a. Briefly explain state of stress at a point. (03 Marks)
- b. Derive the equilibrium equations for three dimensional stress systems. (08 Marks)
- c. Consider the displacement field  $u = [y^2i + 3yzj + (4 + 6x^2)k] \times 10^{-2}$ , what are the rectangular strain components at the point P(1, 0, 2)? Use only linear terms. (05 Marks)

OR

- 8 a. Explain statically determinate and indeterminate structure. (04 Marks)
- b. Using the method of joints, determine the forces in all members of truss shown in Fig. Q8(b). (12 Marks)

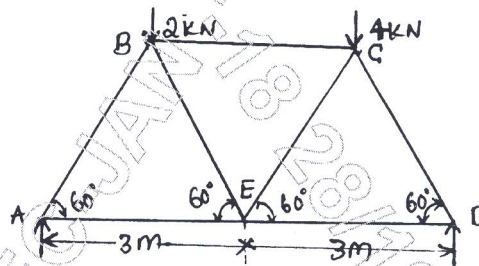


Fig Q8(b)

Module-5

- 9 a. What is strain energy? Derive the equation for strain energy due to bending and torsion. (09 Marks)
- b. State and explain Castiglino's theorem. Using Castiglino's theorem find the deflection of a cantilever subjected to point load P at its free end. (07 Marks)

OR

- 10 a. State the assumptions and explain the limitations of Euler's theory. (04 Marks)
- b. Formulate Rankine-Gordon Equation. (04 Marks)
- c. A 2.5m long hollow circular column with inner diameter to outer diameter ratio 0.8 is to carry a load of 136kN. One end of the column is fixed and the other end is hinged. Determine the diameters of the column. Take  $\sigma_c = 320\text{MPa}$ ,  $a = \frac{1}{7500}$  for material of column, FOS = 2.5. (08Marks)

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